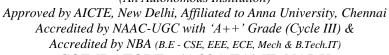
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COIMBATORE-641 035, TAMIL NADU

DEPARTMENT OF AEROSPACE ENGINEERING

Faculty Name : Dr.M.Subramanian,

Academic Year

2024-2025 (Odd)

Year & Branch

Prof & Head/ Aerospace

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T7

Tour & Brunen

III Aerospace

Semester

 \mathbf{V}

Course

19ASB302 – Finite Element Method for Aerospace

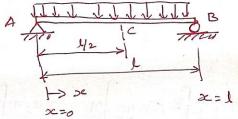
Unit: 1

using trigonometric social - two term total function

Derive the expression for deflection and bending moment in a simply supported beam of span length I, subjects 15 uniformly distributed load over entire span using two term trigonometric troial function.

Also find the deflection and moment at mid span and compare with exact for solution.

Solution. Use Rayleigh-Ritz weltood.



Two term trigonometrie train function suchas $y = a_1 \sin \frac{\pi}{2} + a_2 \sin \frac{3\pi}{2} ---- 0$

The boundary condition for the uniformly distributed loaded beam are

y=0 at x=0 and at x=1

Since the above Solected toral function satisfies the boundary conditions, this function is Correct.

No applying the total potential energy concept we get,

TT = W - W

where U = Internal Strain energy W = workdone by the external force





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III Aerospace Semester

Course : 19ASB302 – Finite Element Method for Aerospace

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Strain energy for a beam is given by

$$V = EI$$
 $V = EI$
 $V = EI$





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Semester : V

Course :

19ASB302 – Finite Element Method for Aerospace

Unit: 1

$$\frac{1}{2\pi} \int_{-\pi}^{\pi} \left[a_1^2 \sin^2 \pi \alpha + 81a_2^2 \sin^2 3\pi \alpha + 12a_1a_2 \sin \pi \alpha \sin \pi \alpha \right]$$
Using the relation (a+b)² = a^2+b^2+aab

Now, the strain energy equation (i'e Eqn @] implies.

$$U = \frac{EI}{2} \int_{-\pi}^{\pi} \left[\frac{d^2y}{dx^2} \right]^2 doc$$

$$= \frac{EI}{2} \int_{\pi}^{\pi} \left[\frac{d^2y}{dx^2} \right]^2 doc$$

$$= \frac{EI}{2} \int_{\pi}^{\pi} \left[\frac{d^2y}{dx^2} \right]^2 doc$$

$$= \frac{EI\pi^4}{2^4} \int_{\pi}^{\pi} \left[a_1^2 \sin^2 \pi \alpha + 81a_2^2 \sin^2 3\pi \alpha + 18a_1a_2 \sin \pi \alpha \sin \pi \alpha + 18a_1a_2 \sin \pi \alpha \cos \pi \alpha \right] doc$$

$$= \frac{EI\pi^4}{2^4} \int_{\pi}^{\pi} \left[a_1^2 \sin^2 \pi \alpha + 81a_2^2 \sin^2 3\pi \alpha + 18a_1a_2 \sin \pi \alpha \cos \pi \alpha \cos \pi \alpha \right] doc$$
Since the Simultaneous integration of all the three inner terms of equation (4) is stightly difficult, let us integral to relation them individually indiv

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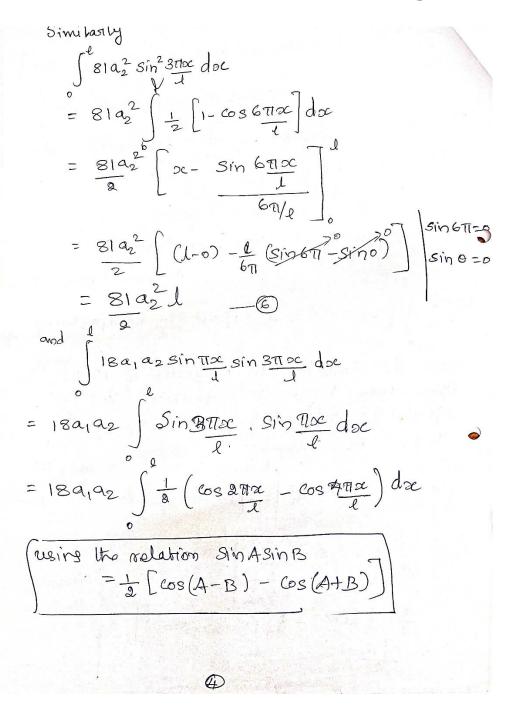
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Prof & Head/ Aerospace
III Aerospace

Semester

 \mathbf{V}

Course

19ASB302 – Finite Element Method for Aerospace

Unit: 1

$$= 9a_1a_2 \left[\frac{\sin 2\pi \alpha}{1} - \frac{\sin 4\pi \alpha}{1} \right]$$

$$= 9a_1a_2 \left[\frac{1}{2\pi} \left(\frac{\sin 2\pi \alpha}{1} - \frac{\sin 4\pi \alpha}{1} \right) - \frac{1}{4\pi} \left(\frac{\sin 4\pi \alpha}{1} - \frac{\sin 4\pi \alpha}{1} \right) \right]$$

$$= 9a_1a_2 \left[\frac{1}{2\pi} \left(\frac{\sin 2\pi \alpha}{1} - \frac{\sin 4\pi \alpha}{1} \right) - \frac{1}{4\pi} \left(\frac{\sin 4\pi \alpha}{1} - \frac{\sin 4\pi \alpha}{1} \right) \right]$$

$$= \frac{1}{2\pi} \left[\frac{1}{2\pi} \left(\frac{\sin 2\pi \alpha}{1} + \frac{1}{2\pi} \right) \right]$$

$$= \frac{1}{2\pi} \left[\frac{1}{2\pi} \left(\frac{1}{2\pi} + \frac{1}{2\pi} \right) \right]$$

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$$= \frac{1}{2\pi} \left[\frac{1}{2\pi} \left(\frac{1}{2\pi} + \frac{1}{2\pi} + \frac{1}{2\pi} \right) \right]$$

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$$= \frac{1}{2\pi} \left[\frac{1}{2\pi} \left(\frac{1}{2\pi} + \frac{1}{2\pi} + \frac{1}{2\pi} + \frac{1}{2\pi} + \frac{1}{2\pi} \right) \right]$$

$$= \frac{1}{2\pi} \left[\frac{1}{2\pi} \left(\frac{1}{2\pi} + \frac{1}{2\pi} + \frac{1}{2\pi} + \frac{1}{2\pi} + \frac{1}{2\pi} + \frac{1$$

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DEPARTMENT OF AEROSPACE ENGINEERING

Faculty Name : Dr.M.Subramanian, Prof & Head/ Aerospace Academic Year : 2024-2025 (Odd)

Year & Branch : III Aerospace Semester : V

Course : 19ASB302 – Finite Element Method for Aerospace

$$= W \left[-\frac{a_1 t}{\Pi} \left(\cos \pi - \cos \theta \right) - \frac{a_2 t}{3 \pi} \left(\cos 3\pi - \cos \theta \right) \right] \frac{\partial}{\partial \theta}$$

$$= W \left[-\frac{a_1 t}{\Pi} \left(-1 - 1 \right) - \frac{a_2 t}{3 \pi} \left(-1 - 1 \right) \right]$$

$$= W \left[\frac{a_1 t}{\Pi} + \frac{a_2 t}{3 \pi} \right]$$

$$= \frac{a_1 t}{\Pi} \left(\frac{a_1 + a_2 t}{3 \pi} \right) - \frac{a_2 t}{\Pi}$$

$$= \frac{a_1 t}{\Pi} \left(\frac{a_1 + a_2 t}{3 \pi} \right) - \frac{a_2 t}{\Pi}$$

$$= \frac{a_1 t}{\Pi} \left(\frac{a_1 + a_2 t}{3 \pi} \right) - \frac{a_2 t}{\Pi} \left(\frac{a_1 + a_2 t}{3 \pi} \right)$$

$$= \frac{a_1 t}{\Pi} \left(\frac{a_1 + a_2 t}{4 \pi} \right) - \frac{a_2 t}{\Pi} \left(\frac{a_1 + a_2 t}{3 \pi} \right)$$

$$= \frac{a_1 t}{\Pi} \left(\frac{a_1 + a_2 t}{4 \pi} \right) - \frac{a_2 t}{\Pi} \left(\frac{a_1 + a_2 t}{3 \pi} \right)$$
For minimum potential energy, the following condition must be satisfied.
$$\frac{\partial \pi}{\partial a_1} = \frac{a_1 \pi}{4 t^3} \left(\frac{a_1 t}{4 t^3} \right) - \frac{a_2 t}{2 \pi} = 0$$

$$\Rightarrow \frac{\partial \pi}{\partial a_2} = \frac{a_1 \pi}{4 t^3} \left(\frac{a_1 t}{4 t^3} \right) - \frac{a_2 t}{2 \pi} = 0$$

$$\Rightarrow \frac{\partial \pi}{\partial a_2} = \frac{a_1 \pi}{4 t^3} \left(\frac{a_2 t}{4 t^3} \right) - \frac{a_2 t}{3 \pi} = 0$$

$$\Rightarrow \frac{a_2 t}{4 \pi} = \frac{a_1 t}{4 t^3} \left(\frac{a_2 t}{4 t^3} \right) - \frac{a_2 t}{3 \pi} = 0$$

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$$\Rightarrow \frac{a_2 t}{4 \pi} = \frac{a_1 t}{4 t^3} \left(\frac{a_2 t}{4 t^3} \right) - \frac{a_2 t}{3 \pi} = 0$$





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Semester

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Course

19ASB302 – Finite Element Method for Aerospace





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Course :

19ASB302 – Finite Element Method for Aerospace

Unit: 1

Determination of bending moment at mid-span too a beam, the bending moment is given by the expression as

Bending moment, NI = EI d 24

Tax 2

From the previous derivation we know that $\frac{d^{2}y}{dx^{2}} = -a_{1}\pi^{2} \sin \pi x - 9a_{2}\pi^{2} \sin 3\pi x$ Hence $M = EI \left(-\frac{a_1 \pi^2}{L^2} \sin \frac{\pi \omega}{J} - \frac{9 a_2 \pi^2}{L^2} \sin \frac{\pi \pi \omega}{J} \right)$ $= - EI \pi^{2} \left(a_{1} Sin \Re x + ga_{2} Sin S \pi x \right)$ In the above equation, we get M=- EITI2 (4WLH Sin TIX + 9x4WLH Sin3TIX)
Et 175 L 243EITI5 L 2 - (EIT) (4 Welf) [Sin ADC + 9 Sin 3TT2] Bending moment at x=1 is obtained as Maximum bending moment $M_{max} = -\frac{4\omega^2}{\pi^3} \left[\sin \left(\frac{9}{4} \times \frac{1}{2} \right) + \frac{9}{348} \sin \left(\frac{37}{4} \times \frac{1}{2} \right) \right]$ $= -4\omega^2 \left[\sin \frac{9}{2} + \frac{9}{343} \sin \frac{37}{2} \right]$

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Course

19ASB302 – Finite Element Method for Aerospace

Unit: 1

$$= -\frac{4\omega \ell^2}{\pi^3} \left(1 - \frac{9}{243} \right) \qquad \left[:: S(n\pi) = 1 \quad \text{S}(n\pi) = -i \right]$$

$$= -3.852 \text{W}^{2}$$

$$= -\text{W}^{2}$$

$$= -\frac{wl^2}{8.05}$$

Negative sign is deedt downward loading