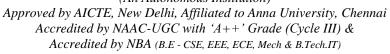


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2024-2025 (Odd)

COIMBATORE-641 035, TAMIL NADU

DEPARTMENT OF AEROSPACE ENGINEERING

Faculty Name : Dr.M.Subramanian, Prof & Head/ Aerospace Academic Year

III Aerospace Semester : V

Course : 19ASB302 – Finite Element Method for Aerospace

Unit: 1

Year & Branch

Find the deflection at the Centre of a Clamped beam Subjected to uniformly distributed load through its length as shown in fig. Use Collocation method and Galarkin's wethod.

A Jammanning B

L d

Differential equation governed by the beam is $EId^2y - W = 0$ $0 \le x \le 1$

Boundary condition's are,

Deflection, y = 0 at 0 = 0 and 0 = 1. Slop, $\theta = E E D = 0$ at 0 = 0 at 0 = 0. Solution

The trial function should Satisfy the given boundary condition. Hence the trail function is considered as.

 $y = c \left(x^5 - 2 l x^4 + l^2 x^3 \right)$

 $\frac{dy}{dsc} = C(5sc^4 - 2(4x^3 + l^2sx^2))$ $= C(5x^4 - 8lx^3 + 3l^2x^2)$

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19ASB302 – Finite Element Method for Aerospace

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The above two functions satisfy the given two boundary anditions. $\frac{d^2y}{dx^2} = C\left(20x^3 - 24 lx^2 + 6l^2x\right)$ $\frac{d^2y}{doc^2} = c \left(60x^2 - 48 le + 6l^2 \right)$ d2y

doct

doct

Substituting the above value in

governing differential equation,

we get the residuals as Rd = CEI (120x-481)-4 Now eve apply, a) allocation point method In this method Ra = 0 " CEI (1200c-481)-W =0 Put oc = 1/2 for maximum deflection . CEI (120x 1/2-481) = W Hence trail function $\mathcal{Y} = \frac{W}{12EEL} \left(x^5 - 2bx^4 + \ell^2 x^2 \right)$





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19ASB302 - Finite Element Method for Aerospace

Unit: 1

At x=1/2, maximum deflection Ymax = W ((1)2- 22(2)4+l2(1)2) Ymac = W14 Which is equal to 384EI Cocact Soluction. ii) Galerkin's method

In this wethood trial function itself
is considered as weighting function
i.e., Twiki die =0 $W_i = C \left[x^5 - 2lx^4 + l^2 x^3 \right]$ P1 = CEE (120,00 - 481] - W Substitude Wi and Rd Value in above equation [c (25-20x4-1-1222) × [CEI(120x-484)-w]/k Integrating above equation and simplifying we get C= W Henco the trail function y= W (25-2lx4+ l2x3) At $x = \frac{l}{2}$ maximum deflection

That $\frac{l}{12EIJ} \left(\frac{l}{2}\right)^5 - 2l\left(\frac{l}{2}\right)^4 + l^2\left(\frac{l}{2}\right)^3$ That $\frac{l}{348EI}$ (3)

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Course

19ASB302 - Finite Element Method for Aerospace

Unit: 1

A fin as shown in fig., governed by a differential quation.

-d [KAGE)d+]+ Ap (T-Tx)=0

Boundary Ondition are

T= 80°c at x = 0

-KAdT? = hA(T-Tx)atx=8cm

ruhan fines end is open to the atmosphere

-KAdT

Doc = 0 at x=8cm (When free end is insulated)

Find the approximate solution when the free end

i) open to atmosphere ii) Insulated

by using Collocation meltod. Assume four

term togal solution.

Take, K= 3 W/cm C 80c 1 Scm + 4cm h = 0.1 W/cm2c

To = 20°C





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19ASB302 - Finite Element Method for Aerospace

Unit: 1

Data Given: Governing equation is given by, -dy [KA(x) of + hp (T-Ta) =0

Boundary Conditions are,

$$T = 80^{\circ}c$$
 at $x = 0$
 $-KAdT$ = $hA(T_{o} - 20)$

Solution:

i) Free end is open to the etmosphere

$$A(sc) = 1 \times 4 = 4 \text{ cm}^2$$

governing equation an be written as,

$$-12\left(\frac{d^2+}{dx^2}\right)+(T-20)=0$$

Assuming a solution

Applying boundary anditions 1 a = 80°C

Applying boundary anditions 2





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Course

19ASB302 - Finite Element Method for Aerospace

Unit: 1

 $3 \times 4 \left(a_1 + 2a_2 \times 4 \cdot 3a_3 \times^2 \right)_{x=8} = A + (T - 20)_{5c=8}$ $12 \left(a_1 + 16a_2 + 192a_3 \right) = 0.1 \times 4 \left(80 + 8a_1 + 16a_2 + 512a_3 \right)$ $3a_1 + 48a_2 + 2504a_3 = 8 + 0.8a_1 + 1.6a_2 + 51.2a_2$

2.201 = 8-46.402 - 2252.803

9, = 3.64-21.192-102493-0

T(x) = 80 + (3.64 - 21.192 - 102492)x $+ 92x^{2} + 92x^{3}$

Bubstituting in governing equation

 $-12 \frac{d^{2}}{dx^{2}} \left[80 + (3.64 - 21.102 - 102403) \right]$ $+ a_{2}c^{2} + a_{3}x^{3}$

+ 80 + (3.64-21.192-102493)2

 $+ 9200^2 + 9300^3$

-> -12 d [3.64-21.1a2-1024a3+2a2x+3a3x3]

+60 + (3.64-21.102-102493)x+92x2+98x3

=R

 $\rightarrow -12(2a_2+6a_3x)+60+(5.64-21.1a_2-10.24a_8)x$ $+a_2x^2+a_3x^3=R(x,a_2,a_3)$

 \Rightarrow $94a_{9}-72a_{3}x+3.64x-21.1a_{2}x-10.24a_{3}x$ $+a_{2}x^{2}+a_{3}x^{3}+60=0(x,a_{2},a_{3})$





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Course

19ASB302 - Finite Element Method for Aerospace

Unit: 1

$$a_2(x^2-21\cdot|x+24) + a_3(x^3-1024x-72x)$$

+60 +364x = R

$$\Rightarrow R(x, a_2, a_3) = 3.64 \times +60^{-1}$$

$$+a_3(x^2-21.10c-24) + a_3(x^3-1024x-72x)$$

By Collocation wethool

-> Ab x=6; R:0

Solving equation (2) and (3)

From equation (1)

91= 4.933

Honce approprimate Solution,





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Course

19ASB302 - Finite Element Method for Aerospace

Unit: 1

(i) If the free end is insulated Foretning equations
- K [d AdT] + Ap (T-Ta) =0 -KAd2T + Ap (T-Too)=0 Boundary Condition, T= 30 at x=0 -KAdT =0 as ezd LOT T(x) = 90 + 91x + 93x2 + 93x3 Applying Boundary condition (1) ao = 80° c Applying Boundary condition (2) (dT) 2=8 = (a, +2a2x+43a3x2) | x=8 =0 $\rightarrow 91 = (-2a_0x - 3a_3x^2)|_{x=8}$ a1 = -16a2 - 19293 T(x) = 80-16a2x-19293x+92x2+93x3 T(x) = 80 + 92 (x2-16x)+93 (x3-192x)





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Course

19ASB302 - Finite Element Method for Aerospace

Unit: 1

Substituting in the governing equation, -KAd2 (20 + 92 (202-16) + 92 (202-192)] + hp [Tex) -Ta] -12 d fa2(22) + 393223 + Ap(T-Tz)-R -12 (2a2+6a) +0.1×10(20+a2(x2-16x) $+ a_3(x^3 - 192x) = R$ $\rightarrow R(x, 92, 93) = 80 + 92(x^2 - 16x - 24)$ + $43 (x^3 - 192x - 72x)$ -> R(x, 92, 93)=80+92 (x2-16x-24) + 93 (x3-2642) By albertion wethod P=0 at x=3 => 80 + (-63) as + (- 765) as = 0 $63a_2 + 765a_3 = 80 \rightarrow (2)$ R=0 at x=6 => 80+ (-84) 92+ (-1368) 93=0 ⇒ 8492 + 1368 a3 = 80 ---(3) Solving equation (1) and (3) $a_2 = 2.20$ 7 Form equation (1), $a_1 = -16a_2 - 192a_3$ $= -16 \times 2.20 + 192 \times 0.076$ Hanco to approximate adultion is Tex)=20-206x+2.202 + 076

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